

Title (en)  
Improved coolant passage structure for rotor blades in a combustion turbine.

Title (de)  
Kühlkanäle für die Schaufeln einer Gasturbine.

Title (fr)  
Canaux de refroidissement pour les aubes d'une turbine à gaz.

Publication  
**EP 0207799 A2 19870107 (EN)**

Application  
**EP 86305182 A 19860703**

Priority  
US 75165785 A 19850703

Abstract (en)  
A combustion turbine rotor blade is provided with an airfoil portion having a plurality of coolant holes extending radially outwardly therethrough. The coolant holes are tapered to a smaller flow cross-section in the radially outward direction to produce more uniform cooling action over the length of the holes over which tapering is provided.

IPC 1-7  
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IPC 8 full level  
**F01D 5/18** (2006.01)

CPC (source: EP)  
**F01D 5/187** (2013.01); **F05D 2250/231** (2013.01); **F05D 2250/232** (2013.01)

Cited by  
WO2017121689A1; EP1820937A3; US2013052035A1; CN113454322A; EP0550184A1; US5413463A; EP2354453A1; EP2743454A1; US2022170376A1; US11905848B2; EP0319758A1; US4893987A; US2010329862A1; JP2011007181A; US8511990B2; EP1541805A1; US6997679B2; US8157527B2; US6994514B2; EP1422383A3; US8511992B2; US9200526B2; WO2004057157A1; US10686199B2; US11489175B2; EP1820937A2; US10408079B2; US10930942B2; US11901591B2; US8506251B2; US8827646B2; US11060195B2; US6824360B2; US6539627B2; US10734661B2

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