

Title (en)
AXIAL GAS TURBINE

Publication
EP 0313826 B1 19920902 (DE)

Application
EP 88115694 A 19880923

Priority
DE 3736836 A 19871030

Abstract (en)
[origin: JPH01151725A] PURPOSE: To improve cooling performance by arranging a blade ring having blades in the last blading stage, while forming a particular ducting guiding the cooling air in the area of the last stage of the turbine for a gas turbine into which a cooling air flow from a compressor is introduced. CONSTITUTION: In an axial flow gas turbine, a cooling air flow from the last stage of a compressor located in the right side of a rim 5 of the first moving blade of the turbine, is circumferentially accelerated by a torsional blade arranged between the compressor and the first turbine stage, passes through a series cooling air holes 15 and further enters into a cooling system. At this time, the cooling air has to be guided to the area of the last blading stage, a through hole 26 is formed in the web 25 of heat accumulation segment 24 arranged in the outer boundary of the rotor, and the cooling air flow passes through the blade root channels 21 of the last moving blade ring 9 and enters into the cooling air blade 27. A conical moving blade 28 is arranged in the inner side of the blade 27 to avoid separation of the cooling air.

IPC 1-7
F01D 5/08; F02C 7/18

IPC 8 full level
F01D 5/08 (2006.01); **F01D 5/14** (2006.01); **F01D 5/18** (2006.01); **F02C 7/18** (2006.01)

CPC (source: EP US)
F01D 5/081 (2013.01 - EP US); **F01D 5/084** (2013.01 - EP US); **F01D 5/145** (2013.01 - EP US)

Cited by
EP2551453A1; EP0636764A1; EP2520764A1; EP3106613A1; EP0447886A1; US5189874A; FR3054855A1; GB2567103A; GB2567103B; US10954795B2; US9382802B2; US9739151B2; WO2018029408A1; WO9947798A1; US6217280B1; US10001061B2; US8277170B2

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