Title (en)

PITCH GUIDANCE SYSTEM

Publication

EP 0480001 A4 19930908 (EN)

Application

EP 91908197 A 19910326

Priority

US 50521490 A 19900404

Abstract (en)

[origin: WO9115827A1] A pitch guidance system for an aircraft utilizes inertially derived pitch information to provide the pilot with information defining the optimum pitch angle for maximum climb during a wind shear condition. The system utilizes a pitch reference modulator (22) that receives a stall warning discrete from a stall warning system to reduce the commanded pitch angle upon the occurrence of a stall warning to reduce the possibility of stalling the aircraft during degraded performance conditions such as tail winds and engine-out conditions. The system utilizes inertially derived pitch information rather than air mass derived angle of attack information to avoid transients in the angle of attack vane signal, and the commanded pitch angle is biased as a function of altitude (70) and vertical speed (58) to optimize the pitch angle for different altitudes and descent rates.

IPC 1-7

G06F 15/50

IPC 8 full level

G05D 1/06 (2006.01)

CPC (source: EP)

G05D 1/0623 (2024.01); **G05D 1/0661** (2024.01)

Citation (search report)

- [X] WO 8909955 A1 19891019 - SUNDSTRAND DATA CONTROL [US]
- [A] US 924059 A 19090608 - GROBE CHARLES W [US]
- [A] EP 0229197 A1 19870722 - BOEING CO [US]
- [A] GB 2134866 A 19840822 - SUNDSTRAND DATA CONTROL
- [A] WO 8704991 A1 19870827 - SUNDSTRAND DATA CONTROL [US]
- See references of WO 9115827A1

Designated contracting state (EPC)

DE FR GB NL

DOCDB simple family (publication)

WO 9115827 A1 19911017; CA 2063823 A1 19911005; EP 0480001 A1 19920415; EP 0480001 A4 19930908

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