

Title (en)

Mounting of a ceramic matrix composite combustion chamber with flexible shrouds

Title (de)

Befestigung einer Turbinenbrennkammer aus keramischem Matrix-Verbundwerkstoff mit flexiblen Verbindungsringen

Title (fr)

Accrochage de chambre de combustion CMC de turbomachine par viroles de liaison souples

Publication

EP 1265030 A1 20021211 (FR)

Application

EP 02291359 A 20020604

Priority

FR 0107375 A 20010606

Abstract (en)

The aircraft gas turbine engine is held in position by a flexible connecting ring which is made of metal in sectors (56,60). One end (56a,60a) of the ring is fixed (50,52) to the combustion chamber and the second end, which forms a flange, is fixed (54,58) to the envelope. The ring can be bolted to the combustion chamber. the connecting ring can be apertured for ventilation.

IPC 1-7

F23R 3/60

IPC 8 full level

F23R 3/42 (2006.01); **F23R 3/50** (2006.01); **F23R 3/60** (2006.01)

CPC (source: EP US)

F23R 3/60 (2013.01 - EP US); **F05B 2230/606** (2013.01 - EP US)

Citation (search report)

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DE FR GB

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