

Title (en)

Mounting of a ceramic matrix composite combustion chamber with flexible shrouds

Title (de)

Befestigung einer Turbinenbrennkammer aus keramischem Matrix-Verbundwerkstoff mit flexiblen Verbindungsringen

Title (fr)

Accrochage de chambre de combustion CMC de turbomachine par viroles de liaison souples

Publication

EP 1265030 B1 20080709 (FR)

Application

EP 02291359 A 20020604

Priority

FR 0107375 A 20010606

Abstract (en)

[origin: EP1265030A1] The aircraft gas turbine engine is held in position by a flexible connecting ring which is made of metal in sectors (56,60). One end (56a,60a) of the ring is fixed (50,52) to the combustion chamber and the second end, which forms a flange, is fixed (54,58) to the envelope. The ring can be bolted to the combustion chamber. the connecting ring can be apertured for ventilation.

IPC 8 full level

F23R 3/42 (2006.01); **F23R 3/60** (2006.01); **F23R 3/50** (2006.01)

CPC (source: EP US)

F23R 3/60 (2013.01 - EP US); **F05B 2230/606** (2013.01 - EP US)

Citation (examination)

- US 5851679 A 19981222 - STOWELL WILLIAM R [US], et al
- US 4739621 A 19880426 - PETTENGILL JASON S [US], et al

Designated contracting state (EPC)

DE FR GB

DOCDB simple family (publication)

EP 1265030 A1 20021211; **EP 1265030 B1 20080709**; DE 60227455 D1 20080821; FR 2825787 A1 20021213; FR 2825787 B1 20040827; JP 2002372242 A 20021226; JP 3984101 B2 20071003; US 2003000223 A1 20030102; US 6823676 B2 20041130

DOCDB simple family (application)

EP 02291359 A 20020604; DE 60227455 T 20020604; FR 0107375 A 20010606; JP 2002158746 A 20020531; US 16180502 A 20020605