

Title (en)
BYPASS TURBOJET ENGINE NACELLE

Title (de)
BYPASSGASTURBINENGONDEL

Title (fr)
NACELLE DE TURBORÉACTEUR À DOUBLE FLUX

Publication
EP 2268910 A2 20110105 (FR)

Application
EP 09742294 A 20090409

Priority
• FR 2009050643 W 20090409
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Abstract (en)
[origin: WO2009136096A2] The invention relates to a bypass turbojet engine nacelle comprising a downstream section equipped with a thrust reverser device comprising a moving cowl (6) mounted such that it can move in translation in a direction substantially parallel to a longitudinal axis of the nacelle and able to move alternately from a closed position in which it ensures the aerodynamic continuity of the nacelle and covers deflection means (70) and an open position in which it opens a passage in the nacelle and uncovers the deflection means, the said moving cowl also being extended by at least one nozzle section (7) mounted at a downstream end of the said moving cowl, characterized in that the nozzle section comprises at least one panel (10) mounted such that it can rotate about at least one pivot about an axis substantially perpendicular to a longitudinal axis of the nacelle, the said panel further being connected to a fixed fairing structure (2) of the turbojet engine by at least one link rod (11) mounted such that it can rotate about anchor points on the panel of the nozzle section and on the fixed structure, respectively.

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F02K 1/12 (2006.01); **F02K 1/72** (2006.01)

CPC (source: EP US)
F02K 1/1261 (2013.01 - EP US); **F02K 1/72** (2013.01 - EP US); **F05D 2260/50** (2013.01 - EP US)

Citation (search report)
See references of WO 2009136096A2

Citation (examination)
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DOCDB simple family (publication)
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