

Title (en)

TRANSITION REGION FOR A SECONDARY COMBUSTION CHAMBER OF A GAS TURBINE

Title (de)

ÜBERGANGSBEREICH FÜR EINE SEKUNDÄRBRENNKAMMER EINER GASTURBINE

Title (fr)

ZONE DE TRANSITION POUR UNE CHAMBRE DE COMBUSTION SECONDAIRE D'UNE TURBINE À GAZ

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Application

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Priority

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Abstract (en)

[origin: WO2011138193A1] The invention relates to a gas turbine having a secondary combustion chamber (1) and a first guide vane row (2) of a low pressure turbine, arranged directly downstream thereof, wherein the radial exterior delimitation of the secondary combustion chamber (1) is formed by at least one outer wall segment (4), which is fastened to at least one carrier element (5), arranged radially on the outside, wherein the flow path of the hot gases (3) is delimited radially on the outside by an outer platform (6) in the region of the guide vane row (2) which is fastened at least indirectly to at least one guide vane carrier (8), and wherein there is a gap-shaped cavity (9), running substantially radially and having a width (B) in the inlet region in the axial direction in the range of 1-25 mm, between the wall segment (4) and the outer platform (6). The gas turbine according to the invention is in particular characterized in that at least one step element (22, 22', 22'') is arranged in the inlet region, which reduces the cited width (B) by at least 10% in at least one step (28), running substantially perpendicularly to the flow direction (11) of the hot gas in the cavity (9).

IPC 8 full level

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Citation (search report)

See references of WO 2011138193A1

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