

Title (en)  
NACELLE FOR AN AIRCRAFT TURBOFAN ENGINE

Title (de)  
TRIEBWERKSGONDEL FÜR EIN TURBOFAN-TRIEBWERK

Title (fr)  
NACELLE POUR UN TURBORÉACTEUR DOUBLE FLUX D'UN AÉRONEF

Publication  
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Application  
**EP 11831835 A 20111028**

Priority

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Abstract (en)  
[origin: WO2012066210A2] The invention relates to a nacelle (1) for an aircraft bypass turbojet engine comprising, in downstream cross section, an inner fixed structure (8) intended to surround part of the bypass turbojet engine and an outer structure (9) at least partially surrounding the inner fixed structure (8) so as to delimit an annular flow path (10), the outer structure (9) comprising at least one inner flap (101) positioned facing the annular flow path (10), an outer flap (103) not in contact with the annular flow path (10) at least partially surmounting each inner flap (101) in aerodynamic continuity with the rest of the outer structure (9), and an intermediate flap (105) positioned between each inner flap (101) and each outer flap (103), said intermediate flap (105) being capable of translational movement so as to increase or decrease the cross section of the annular flow path (10), and each inner flap (101) and each outer flap (103) being capable of rotational movement so as to remain in permanent contact with the intermediate flap (105) in all positions of the latter.

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