

Title (en)
HYBRIDISATION OF THE COMPRESSORS OF A TURBOJET

Title (de)
HYBRIDISIERUNG DER VERDICHTER EINER STRAHLTURBINE

Title (fr)
HYBRIDATION DES COMPRESSEURS D'UN TURBORÉACTEUR

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Application
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Abstract (en)
[origin: WO2016020618A1] The present invention relates to a twin-flow, double body turbojet comprising a fan (S) positioned upstream from a gas generator and delimiting a primary flow and a secondary flow, said gas generator being traversed by the primary flow and comprising a low-pressure compressor (1), a high-pressure compressor (2), a combustion chamber (3), a high-pressure turbine (4) and a low-pressure turbine (5), said low-pressure turbine being linked to said low-pressure compressor by a low-pressure rotating shaft (10) and said high-pressure turbine being linked to said high-pressure compressor by a high-pressure rotating shaft (20), characterised by the fact that said turbojet comprises an electric motor forming a device for injecting mechanical power (8) into at least one of said rotating shafts (10, 20), a device (7) for removing power from at least one of said rotating shafts, dimensioned to extract excess power (w3, w5) relative to the requirement for actuating accessories of the turbojet (w7), transforming said excess power into electrical energy, and an electric storage means (9) positioned between said device for removing power and said electric motor.

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Opponent : Raytheon Technologies Corporation
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