

Title (en)
TURBOMACHINE COMBUSTION CHAMBER

Title (de)
BRENNKAMMER EINER TURBOMASCHINE

Title (fr)
CHAMBRE DE COMBUSTION DE TURBOMACHINE

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Application
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Abstract (en)
[origin: WO2016051067A1] The invention relates to a combustion chamber for a turbomachine, such as an aircraft turbojet engine or turboprop engine, comprising an internal annular shroud and an external annular shroud which at their upstream ends are connected by an annular chamber end wall (48), said chamber comprising deflectors (50) mounted upstream of the annular chamber end wall (48). Injectors (19) are mounted in sleeves (44) at least one of which comprises a radially annular flange (66) which is designed to slide radially between the chamber end wall (48) and the deflector (50) and which is blocked axially between the chamber end wall (48) and the deflector (50).

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