

Title (en)

FILM COOLING HOLE IN GAS TURBINE COMPONENTS

Title (de)

FILMKÜHLLOCH IN GASTURBINEN BAUTEILEN

Title (fr)

TROU POUR FILM D'AIR DE REFROIDISSEMENT DANS DES PIÈCES DE TURBINE À GAZ

Publication

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Application

EP 17715064 A 20170322

Priority

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- EP 2017056834 W 20170322

Abstract (en)

[origin: WO2017162743A1] The invention relates to a film cooling hole (20) of gas turbine components to be cooled, having an inflow section (26) with a constant flow area, to which a diffuser section (28) with a variable flow area is connected. The aim of the invention is to provide a particularly efficient film cooling. To this end, the diffuser region (26) is simply widened in the direction perpendicular to the flow direction of the hotter medium MH.

IPC 8 full level

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CPC (source: EP US)

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Citation (search report)

See references of WO 2017162743A1

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