

Title (en)  
DIRECTIONAL COOLING ARRANGEMENT FOR TURBINE AIRFOILS

Title (de)  
DIREKTIONALE KÜHLANORDNUNG FÜR TURBINENSCHAUFEL

Title (fr)  
AGENCEMENT DE REFROIDISSEMENT DIRECTIONNEL DESTINÉ À DES AUBES DE TURBINE

Publication  
**EP 3444436 B1 20201223 (EN)**

Application  
**EP 18189576 A 20180817**

Priority  
US 201715679442 A 20170817

Abstract (en)  
[origin: EP3444436A1] An airfoil (61; 161; 262; 361) includes an internal wall (189; 289; 389) and an external wall (188; 288; 388). The external wall (188; 288; 388) defines pressure and suction sides (P, S) between a leading edge (66; 166; 266; 366) and a trailing edge (68; 168), and the airfoil section (65; 165) defines a mean camber line(75; 175; 275; 375) that extends between the leading and trailing edges (66, 68; 166, 168; 266; 366) to bisect a thickness of the airfoil section(65; 165). A first cavity (184; 284; 384) and a second cavity (186; 286; 386) are separated by the internal wall (189; 289; 389). The second cavity (186; 286; 386) is bounded by the external wall (188; 288; 388) at the leading edge (66; 166; 266; 366). At least one crossover passage (194; 294; 394) within the internal wall (189; 289; 389) connects the first cavity (184; 284; 384) to the second cavity (186; 286; 386). The crossover passage (194; 294; 394) defines a passage axis (PA). The passage axis (PA) defines a passage angle ( $\alpha$ ) with respect to the mean camber line (75; 175; 275; 375) such that the passage axis (PA) extends transversely from the mean camber line (75; 175; 275; 375) to intersect a surface of the second cavity (186; 286; 386).

IPC 8 full level  
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