

Title (en)

AIRFOIL HAVING IMPROVED LEADING EDGE COOLING SCHEME AND DAMAGE RESISTANCE

Title (de)

SCHAUFELBLATT MIT VERBESSERTEM VORDERKANTENKÜHLSHEMA UND BESCHÄDIGUNGSWIDERSTAND

Title (fr)

PROFIL AÉRODYNAMIQUE PRÉSENTANT UN SCHÉMA AMÉLIORÉ DE REFROIDISSEMENT DE BORD D'ATTAQUE ET UNE MEILLEURE RÉSISTANCE AUX DOMMAGES

Publication

EP 3567218 B1 20220803 (EN)

Application

EP 19172115 A 20190501

Priority

US 201815972637 A 20180507

Abstract (en)

[origin: EP3567218A1] Airfoil for a gas turbine including a body (402) extending between leading (412) and trailing edges (414) in an axial direction, between pressure (416) and suction sides (418) in a circumferential direction, and between a root (406) and tip (408) in a radial direction. A first transitioning leading edge cavity (422) is located adjacent one of the sides proximate the root of the body and transitions axially toward the leading edge as the first transitioning leading edge cavity extends radially toward the tip. A second transitioning leading edge cavity (424) is adjacent the other side and adjacent the leading edge proximate the root of the body and transitions axially toward the trailing edge as the second transitioning leading edge cavity extends radially toward the tip. A portion of the second transitioning leading edge cavity shields a portion of the first transitioning leading edge cavity proximate the root of the body.

IPC 8 full level

F01D 5/18 (2006.01)

CPC (source: EP US)

F01D 5/187 (2013.01 - EP US); **F01D 9/041** (2013.01 - US); **F01D 25/12** (2013.01 - US); **F01D 5/186** (2013.01 - US); **F05D 2220/32** (2013.01 - US); **F05D 2240/121** (2013.01 - US); **F05D 2240/303** (2013.01 - EP US); **F05D 2260/201** (2013.01 - US); **F05D 2260/202** (2013.01 - US)

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