

Title (en)

TURBINE ENGINE COMBUSTION CHAMBER

Title (de)

BRENNKAMMER FÜR EINEN TURBINENMOTOR

Title (fr)

CHAMBRE DE COMBUSTION POUR TURBOMACHINE

Publication

EP 3568638 A1 20191120 (FR)

Application

EP 18700941 A 20180104

Priority

- FR 1750208 A 20170110
- FR 2018050021 W 20180104

Abstract (en)

[origin: WO2018130765A1] The invention relates to a combustion chamber (6) for a turbine engine, in particular for an aircraft turbojet engine or turboprop engine, comprising: - a radially outer annular shroud (11), - a radially inner annular shroud (12), coaxial with the radially outer shroud (11), - an end wall (13) connecting the radially outer shroud (11) and the radially inner shroud (12), characterised in that it comprises a first annular sealing member (38a), coaxial with the radially inner (12) and outer (11) shrouds, the first sealing member (38a) being radially interposed between the end wall (13) and the radially outer shroud (11).

IPC 8 full level

F23R 3/60 (2006.01)

CPC (source: EP US)

F23R 3/002 (2013.01 - US); **F23R 3/60** (2013.01 - EP US); **F23R 2900/00012** (2013.01 - EP US)

Citation (search report)

See references of WO 2018130765A1

Designated contracting state (EPC)

AL AT BE BG CH CY CZ DE DK EE ES FI FR GB GR HR HU IE IS IT LI LT LU LV MC MK MT NL NO PL PT RO RS SE SI SK SM TR

Designated extension state (EPC)

BA ME

DOCDB simple family (publication)

FR 3061761 A1 20180713; **FR 3061761 B1 20210101**; CN 110168284 A 20190823; CN 110168284 B 20210223; EP 3568638 A1 20191120; EP 3568638 B1 20210331; US 11614234 B2 20230328; US 2019360698 A1 20191128; WO 2018130765 A1 20180719

DOCDB simple family (application)

FR 1750208 A 20170110; CN 201880006047 A 20180104; EP 18700941 A 20180104; FR 2018050021 W 20180104; US 201816476776 A 20180104