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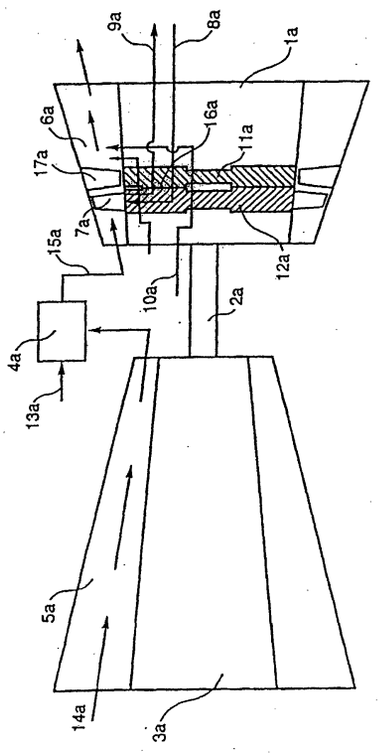
(54) Coolant recovery type gas turbine

(57) A gas turbine includes a rotor shaft comprising a plurality of discs (8, 9, 10, 11, 12a). Each disc (8, 9, 10, 11, 12a) has a plurality of moving blades (4, 5, 6, 7, 7a) driven by combustion gas and arranged annularly on the peripheral portion. Spacers (11a, 13, 14, 15) are arranged between the discs (8, 9, 10, 11, 12a), the respective discs (8, 9, 10, 11, 12a) and the spacers (11a, 13, 14, 15) being arranged in the axial direction. The moving blades (4, 5, 6, 7, 7a) are provided with flow paths (8a, 9a) for introducing coolant for cooling and discharging the coolant heated by the combustion gas. Contact surfaces (16a, 31, 32, 33, 34, 35, 36) contacting both the discs (8, 9, 10, 11, 12a) in rotor peripheral side regions and adjacent spacers (11a, 13, 14, 15) are formed therebetween. A supply flow path (8a) passes through the discs (8, 9, 10, 11, 12a) and the spacers (11a, 13, 14, 15) in the regions forming the contact surfaces (16a, 31, 32, 33, 34, 35, 36) and supply the coolant for cooling the moving blades (4, 5, 6, 7, 7a). A recovery flow path (9a) recovers the coolant which is heated by the moving blades (4, 5, 6, 7, 7a).

FIG. 1



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EUROPEAN SEARCH REPORT

Application Number
EP 03 00 4315

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			F01D

The present search report has been drawn up for all claims

Place of search	Date of completion of the search	Examiner
THE HAGUE	11 September 2003	Argentini, A

CATEGORY OF CITED DOCUMENTS

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