



Europäisches Patentamt
European Patent Office
Office européen des brevets



(11) **EP 1 467 066 A8**

(12) **CORRECTED EUROPEAN PATENT APPLICATION**

Note: Bibliography reflects the latest situation

(15) Correction information:
Corrected version no 1 (W1 A2)
INID code(s) 72

(51) Int Cl.7: **F01D 11/00**, F01D 11/08,
F01D 11/14

(48) Corrigendum issued on:
22.12.2004 Bulletin 2004/52

(43) Date of publication:
13.10.2004 Bulletin 2004/42

(21) Application number: **04251632.8**

(22) Date of filing: **22.03.2004**

(84) Designated Contracting States:
AT BE BG CH CY CZ DE DK EE ES FI FR GB GR
HU IE IT LI LU MC NL PL PT RO SE SI SK TR
Designated Extension States:
AL HR LT LV MK

(30) Priority: **09.04.2003 GB 0308147**

(71) Applicant: **ROLLS-ROYCE PLC**
London, SW1E 6AT (GB)

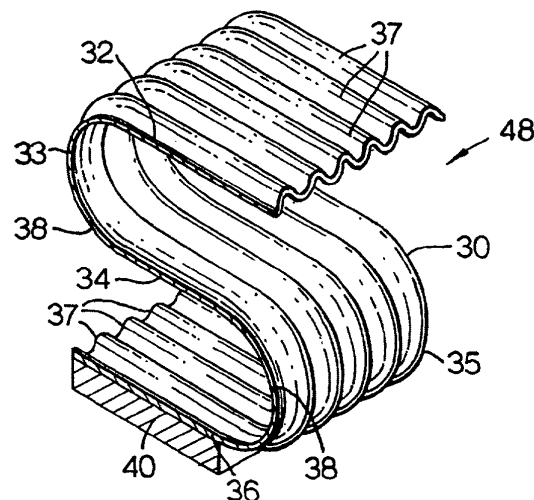
(72) Inventors:
• **Webster, John Richard**
Derby DE23 6AF (GB)
• **Razzell, Anthony Gordon**
Aston on Trent, Derby DE72 2BA (GB)
• **Brown, Philip David**
Oakwood, Derby DE21 2BW (GB)

(74) Representative: **Gunn, Michael Alan**
Rolls-Royce plc
P.O. Box 31
Derby DE24 8BJ (GB)

(54) **Corrugated seal**

(57) A turbofan gas turbine engine (10) comprises a compressor rotor (24) carrying a plurality of circumferentially spaced radially extending rotor blades (26) and a casing (28) surrounding the compressor rotor (24) and compressor rotor blades (26). A compressor rotor blade tip seal (48) comprises an annular member (30) secured to and arranged within the casing (28). The annular member (40) has a plurality of circumferentially spaced axially extending corrugations (32) and a plurality of radially spaced circumferentially extending corrugations (34). A lining (40) is secured to and arranged within the annular member (30). The lining (40) is spaced radially from the tips (27) of the compressor rotor blades (26) to form a clearance (29). An actuator (56) is provided to move the annular member (30) and lining (40) radially to control the size of the clearance (29).

Fig.3.



EP 1 467 066 A8