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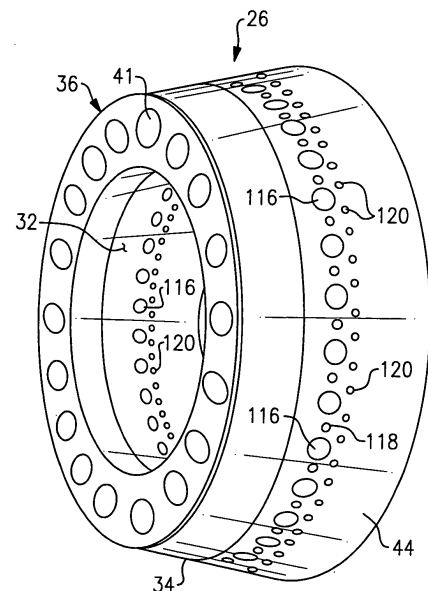
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(54) **Gas turbine combustor with improved quench holes arrangement**

(57) A gas turbine engine has a combustor module including an annular combustor (26) having a liner assembly that defines an annular combustion chamber (30) and includes a circumferential row of a plurality of relatively large combustion dilution air admission holes (116) and a circumferential row of a plurality of smaller quench air admission holes (120) disposed downstream with respect to the flow of combustion gas products. The plurality of quench air admission holes (120) are arranged with respect to the plurality of relatively large dilution air admission holes (116) disposed upstream thereof such that there is associated with each dilution air admission hole (116) a first quench air admission hole (120) and a second quench air admission hole (120), the first quench air admission hole (120) being offset laterally in a first lateral direction and the second quench air admission hole (120) being offset laterally in a second lateral direction opposite to the first direction.

FIG.3





EUROPEAN SEARCH REPORT

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DOCUMENTS CONSIDERED TO BE RELEVANT			
Category	Citation of document with indication, where appropriate, of relevant passages	Relevant to claim	CLASSIFICATION OF THE APPLICATION (IPC)
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			TECHNICAL FIELDS SEARCHED (IPC)
			F23R
The present search report has been drawn up for all claims			
Place of search The Hague		Date of completion of the search 4 September 2014	Examiner Villar Fernández, R
CATEGORY OF CITED DOCUMENTS X : particularly relevant if taken alone Y : particularly relevant if combined with another document of the same category A : technological background O : non-written disclosure P : intermediate document		T : theory or principle underlying the invention E : earlier patent document, but published on, or after the filing date D : document cited in the application L : document cited for other reasons & : member of the same patent family, corresponding document	

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ANNEX TO THE EUROPEAN SEARCH REPORT
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