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(54) **Diffuser for gas turbine system between compressor exit and combustor inlet**

(57) A diffuser (20) for a gas turbine system (10) having a longitudinal axis (90) is disclosed. The diffuser (20) includes a plurality of diffuser ducts (30). Each of the plurality of diffuser ducts (30) is disposed annularly about the longitudinal axis (90) and has an inlet (32), an outlet

(36), and a passage (34) extending between the inlet (32) and the outlet (36). The outlet (36) of each of the plurality of diffuser ducts (30) is tangentially offset from the inlet (32) of the respective diffuser duct (30). Each of the plurality of diffuser ducts (30) is configured to flow a gas flow (80) therethrough, reducing the gas flow (80) velocity.

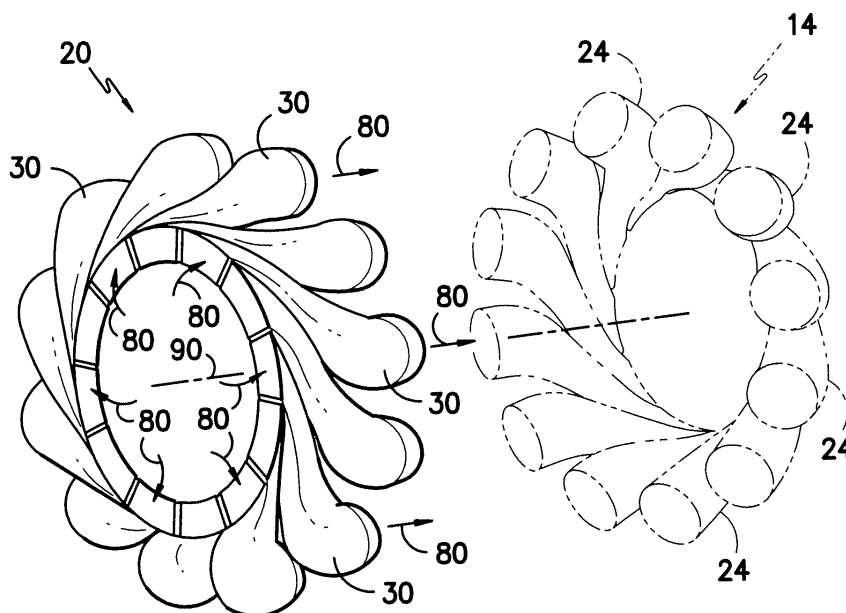


FIG. -3-



EUROPEAN SEARCH REPORT

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EP 11 16 4824

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The present search report has been drawn up for all claims			
Place of search The Hague		Date of completion of the search 26 April 2013	Examiner de la Loma, Andrés
CATEGORY OF CITED DOCUMENTS X : particularly relevant if taken alone Y : particularly relevant if combined with another document of the same category A : technological background O : non-written disclosure P : intermediate document T : theory or principle underlying the invention E : earlier patent document, but published on, or after the filing date D : document cited in the application L : document cited for other reasons & : member of the same patent family, corresponding document			

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